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*Osaka University*
New Interaction Equation for Plate Buckling†

Yukio UEDA*, Sherif M.H. RASHED** and Jeom Kee PAIK***

Abstract

This paper is concerned with proposal of an accurate elastic buckling interaction equation for simply supported rectangular plates subjected to five load components such as compression and bending in two directions and shear. In order to construct this interaction, data on buckling strength of plate under two load components partly calculated in the present study and partly obtained from existing ones are first taken, and then buckling interaction relationships for any two load components are developed. Based on these relationships, the new buckling interaction equation for five load components is theoretically derived. Assessment of accuracy is performed and it is found that the proposed equation has sufficient accuracy for practical designs. Some comparisons are also made with the interactions proposed by Lloyd’s Register and Det Norske Veritas. The result indicates that the new interaction equation proposed in this paper yields better accuracy in safety side.

KEY WORDS: (Elastic Buckling) (Rectangular Plate) (Buckling Interaction Equation)

1. Introduction

Such many plate structures as ships are composed of plate elements, which may sustain mainly in-plane loads and therefore are designed to have sufficient in-plane stiffness and strength.

In these structures, plate buckling is one of the most important design criteria and buckling load may usually be obtained as an eigen-value solution of the governing equations for the plate. However, in many cases, it is very difficult to gain an exact solution for such complicated buckled shape as under any combinations of loading and different boundary condition. Even approximate solutions tend to require a considerable analytical and/or numerical efforts.

To estimate buckling strength for practical design, simple and accurate buckling interaction equations are of great interest to designers, and therefore, charts or simplified equations based on these solutions, included in many handbooks1), have been proposed. However, from the viewpoint of safety, high level accuracy has also been required in evaluation of buckling strength in the design stage.

In this paper, a new and accurate elastic buckling interaction equation is proposed for simply supported rectang-
energy is used. When a rectangular plate under in-plane loads as shown in Fig. 1, buckles, total potential energy $\Pi$ due to lateral deflection $w$ may be obtained as follows:

$$\Pi = U - V$$

(1)

where $U =$ strain energy stored in the plate due to buckling deformation

$$= \int \frac{D}{2} \left[ (\frac{\partial^2 w}{\partial x^2})^2 + 2(1 - \nu)(\frac{\partial^2 w}{\partial x \partial y})^2 \right. $$

$$\left. + 2\nu(\frac{\partial^2 w}{\partial x^2})(\frac{\partial^2 w}{\partial y^2}) + (\frac{\partial^2 w}{\partial y^2})^2 \right] dx dy$$

$V =$ loss of potential energy of external loads due to buckling deformation

$$= \int \frac{t}{2} \left[ \sigma_x(\frac{\partial w}{\partial x})^2 + 2\tau(\frac{\partial w}{\partial x})(\frac{\partial w}{\partial y}) \right. $$

$$\left. + \sigma_y(\frac{\partial w}{\partial y})^2 \right] dx dy$$

and $D = \frac{E t^3}{12(1 - \nu^2)}$, $\nu =$ Poisson’s ratio, $t =$ plate thickness, $E =$ modulus of elasticity and also $\sigma_x, \sigma_y$ and $\tau (= \tau_{xy})$ are stresses loaded at plate edges as shown in Fig. 1. It is the entire volume of the plate that should be integrated by the above equation. The principle of minimum potential energy gives the following buckling criterion:

$$\delta \Pi = 0$$

(2)

For simply supported rectangular plates, the following deflection function which is the form of a Fourier series may be assumed;

$$w = \sum_{m=1}^{\infty} \sum_{n=1}^{\infty} A_{mn} \sin \frac{m\pi x}{a} \sin \frac{n\pi y}{b}$$

(3)

where

$$a, b =$ length and breadth of the plate, respectively

$$A_{mn} =$ coefficients

After substituting Eq.(3) into Eq.(2) and integrating it, the buckling load is determined from the following stationary condition, which represents simultaneous equations.

$$\frac{\partial \Pi}{\partial A_{mn}} = 0 \text{ (m, n = 1 ~ \infty)}$$

(2*)

If the assumed deflection function is the exact one, the final solution is the same as one obtained from the solution of the differential equation. However, in general, it is very difficult to solve the homogeneous equation Eq. (2*) analytically. Therefore, numerical procedures are usually applied.

2.2 Incremental Galerkin method

When buckling load as an eigen-value is calculated by the previous method, it is generally necessary to take a large number of terms of the assumed deflection function in order to gain sufficient accuracy for such complicated shape of buckling as with shear buckling.

As for numerical techniques, they also require considerable computing time. These problems may also occur when post-buckling behaviour of plates with initial deflections is analyzed.

From the above viewpoints, the authors have developed the incremental Galerkin's method for analysis of large deflection behaviour of simply supported rectangular plates under in-plane loads. In this paper, this method is applied to establish non-existing data of buckling strength.

Large deflection theory of plates with initial deflection in the elastic range is governed by the following fundamental equilibrium and compatibility equations;

$$F^w = \frac{t}{D} \left[ \frac{\partial^2 F}{\partial y^2} \frac{\partial^2 (w + w_0)}{\partial x^2} + \frac{\partial^2 F}{\partial x^2} \frac{\partial^2 (w + w_0)}{\partial y^2} \right. $$

$$\left. - 2 \frac{\partial^2 F}{\partial x \partial y} \frac{\partial^2 (w + w_0)}{\partial x \partial y} \right]$$

(4)

$$P^w = E \left[ \left( \frac{\partial^2 w}{\partial x \partial y} \right)^2 - \frac{\partial^2 w}{\partial x^2} \frac{\partial^2 w}{\partial y^2} + 2 \frac{\partial^2 w}{\partial x \partial y} \frac{\partial^2 w}{\partial x \partial y} \right. $$

$$\left. - \frac{\partial^2 w_0}{\partial x^2} \frac{\partial^2 w}{\partial y^2} - \frac{\partial^2 w}{\partial x^2} \frac{\partial^2 w_0}{\partial y^2} \right]$$

(5)

where, $F$ is a stress function satisfying the following relationships;

$$\sigma_x = \frac{\partial^2 F}{\partial y^2}, \sigma_y = \frac{\partial^2 F}{\partial x^2}, \tau = - \frac{\partial^2 F}{\partial x \partial y}$$

and $w =$ deflection due to external load, $w_0 =$ initial
deflection.

Equations (4) and (5) are nonlinear partial differential equations with regard to \( F \) and \( w \), so that it is very difficult to solve the above equations directly when the plate buckles with complicated buckling shape.

Therefore, the fundamental equations are linearized by the incremental technique, as follows:

At the \( (n) \)th step of load increments, deflection and stress function may be expressed as follows:

\[
\begin{align*}
    w &= w^{n-1} + \Delta w \\
    F &= F^{n-1} + \Delta F
\end{align*}
\]

where,

\( w, F = \text{total deflection and stress function for loading from zero to the } (n) \text{th step of load increments, respectively} \)

\( w^{n-1}, F^{n-1} = \text{total deflection and stress function for loading from zero to the } (n-1) \text{th step of load increments, respectively} \)

\( \Delta w, \Delta F = \text{increments of deflection and stress function due to load increment in the } (n) \text{th loading step, respectively} \).

Substituting Eq.(6) into Eqs.(4) and (5), and neglecting the terms of \( \Delta^2 w \) and \( \Delta^2 F \) of higher order than the 2nd order which are negligible when increment of load and deflection is very small, the following fundamental equations which are linear in an interval of a load increment are obtained:

\[
\begin{align*}
    \mathbf{P}^{4}(\Delta w) &= \frac{t}{D} \left[ \frac{\partial^2 \Delta F}{\partial y^2} \frac{\partial^2 w_T}{\partial x^2} + \frac{\partial^2 \Delta F}{\partial x^2} \frac{\partial^2 w_T}{\partial y^2} 
    + \frac{\partial^2 F^{n-1}}{\partial y^2} \frac{\partial^2 \Delta w}{\partial x^2} + \frac{\partial^2 F^{n-1}}{\partial x^2} \frac{\partial^2 \Delta w}{\partial y^2} 
    - 2 \frac{\partial^2 F^{n-1}}{\partial x \partial y} \frac{\partial^2 w_T}{\partial x \partial y} - 2 \frac{\partial^2 F^{n-1}}{\partial x \partial y} \frac{\partial^2 \Delta w}{\partial x \partial y} \right] \\
    \mathbf{P}^{4}(\Delta F) &= E \left[ 2 \frac{\partial^2 w_T}{\partial x \partial y} \frac{\partial^2 \Delta w}{\partial x \partial y} - \frac{\partial^2 w_T}{\partial x^2} \frac{\partial^2 \Delta w}{\partial y^2} \right]
\end{align*}
\]

where,

\( w_T = w^{n-1} + w_0 \)

Assuming \( \Delta w = \sum \sum A_{mn} \Phi_m(x) \cdot \Psi_n(y) \) as a deflection function which satisfies the boundary conditions and substituting this assumed deflection function into Eq.(8), \( \Delta F \) can be expressed as a function of \( \Delta w \). After substituting this obtained \( \Delta F \) and \( \Delta W \) into Eq.(7), \( \Delta w \) and \( \Delta F \) can be obtained without difficulty by the Galerkin method. Ultimately, \( w \) and \( F \) are calculated as follows:

\[
\begin{align*}
    w &= \sum_{i=1}^{n} \Delta w_i \\
    F &= \sum_{i=1}^{n} \Delta F_i
\end{align*}
\]

From the above, for simply supported rectangular plates with initial deflections, it may be assumed that shapes of deflection due to loading and initial deflection are:

\[
\begin{align*}
    w &= \sum_{m} \sum_{n} A_{mn} \sin \frac{m \pi}{a} x \sin \frac{n \pi}{b} y \\
    w_0 &= \sum_{m} \sum_{n} A_{0mn} \sin \frac{m \pi}{a} x \sin \frac{n \pi}{b} y
\end{align*}
\]

and the boundary conditions are:

\[
\begin{align*}
    w &= 0, \frac{\partial^2 w}{\partial x^2} + \nu \frac{\partial^2 w}{\partial y^2} = 0 \text{ at } x = 0, a \\
    w &= 0, \frac{\partial^2 w}{\partial y^2} + \nu \frac{\partial^2 w}{\partial x^2} = 0 \text{ at } y = 0, b \\
    \int_{0}^{l} \sigma_x \, dy &= P_x \\
    \int_{0}^{l} \sigma_y \, dx &= P_y \\
    \int_{0}^{b} \sigma_y \, dx &+ M_x = 0 \\
    \int_{0}^{a} \sigma_y \, dx &+ M_y = 0
\end{align*}
\]

In this analysis, buckling load is obtained not as an eigen-value but as the load applied at the point when deflection in a plate with very small initial deflection increases rapidly as external load is applied incrementally.

The accuracy of the solution also depends on the number of terms taken in deflection series in Eq.(9) and how accurately they can express the buckling shape. However, much more terms may be taken in this method since the linearized fundamental equation gives linear homogeneous equations that are solved directly without difficulty.

Furthermore, buckling load can easily be evaluated with good accuracy if small initial deflection is taken.

If a larger number of terms of deflection series are taken, it converges from upper bound value to exact one.

2.3 Finite element method

According to the theorem of minimum potential energy mentioned in sec. 2.1, the fundamental equation for determination of buckling load by FEM can be derived: A rectangular plate is subdivided into finite elements and deflection of the total plate due to buckling is expressed as an assembly of the deflection of each element;
that is,

$$ w = [A_b] \{w_n\} \quad (11) $$

where,

$$ [A_b] = \text{shape function} \quad \{w_n\} = \text{nodal displacement} $$

After substituting Eq.(11) into Eq.(1), partial differentiation with respect to the nodal displacements is performed, leading to stationary conditions;

$$ 0 = (\Sigma [K_{bb}] + \Sigma [K_I]) \{w_n\} \quad (12) $$

where,

$$ [K_{bb}] = \text{bending stiffness matrix of a finite element} \quad [K_I] = \text{stability coefficient matrix of a finite element,} $$
which is a linear function of stresses

$$ \Sigma = \text{assembly of all elements} $$

Equation (12) has a trivial solution \{w_n\} = 0. Non-trivial solution exists only if the following equation is satisfied;

$$ [\Sigma [K_{bb}] + \Sigma [K_I]] = 0 \quad (13) $$

In the above procedure, when a combination of stresses associated with \([K_I]\) is reached such that, Eq.(13) is satisfied the buckling load is determined as an eigen-value.

The accuracy of buckling load determined by this method depends on the assumed shape function, the number of element and the complexity of the buckling shape.

3. New Buckling Interaction Equation

In this paper, a simply supported rectangular plate which has \(a, b\) and \(t\) that are length, breadth and thickness of the plate respectively, is considered as shown in Fig. 1. When the plate is subjected to combined in-plane loads, such as uniform compression and bending in two directions and uniform shear, a new buckling interaction equation is derived.

3.1 Buckling interaction equation for two load components

As the first step for constructing a new interaction equation, the interaction between each pair of load components such as \(\sigma_x\) and \(\sigma_y\), \(\sigma_{bx}\) and \(\sigma_{by}\), etc. is considered.

Figure 2 to 11 represent these interaction relationships for different values of aspect ratio of the plate. The \(x\) axis can always be chosen parallel to the longer side of the plate, so that \(a/b\) becomes greater than 1.0 and it is unnecessary to consider plates with aspect ratio \(a/b\) smaller than 1.0 in constructing interaction relationships.

Interaction relationships presented in Refs. 1, 5, 7 and 8 are reproduced in these figures except Figs. 5 and 7, which are constructed using the incremental Galerkin method, and Fig. 11, which is partly calculated by the finite element method. The sources of these figures are shown in Table 1.

Since the accuracy of these buckling loads expressed by solid lines in Figs. 2 to 11 is sufficient, they are considered as reference solutions for comparison of accuracy with buckling load obtained by different formulae.

The interaction equations formulated in this paper are also plotted in Figs. 2 to 11. Subscript \(cr\) in the following interaction equations means buckling strength of plates under a single load (see appendix).

**Table 1 Sources of elastic buckling interaction relationships between different load components**

<table>
<thead>
<tr>
<th>(\sigma_x)</th>
<th>(\sigma_y)</th>
<th>(\sigma_{bx})</th>
<th>(\sigma_{by})</th>
<th>(t)</th>
</tr>
</thead>
<tbody>
<tr>
<td>(\sigma_{x1})</td>
<td>(\sigma_{y1})</td>
<td>(\sigma_{bx1})</td>
<td>(\sigma_{by1})</td>
<td>(t)</td>
</tr>
</tbody>
</table>

(1) Biaxial compressions, \(\sigma_x\) and \(\sigma_y\)

Figure 2 shows the buckling interaction relationships of plates with different aspect ratios subjected to biaxial compressions. These relations are obtained by the energy method. Only one term of the deflection series, Eq.(3), in each direction is necessary and the solution in this case is exact.

The interaction relationship is represented by a straight line for \(1 \leq a/b \leq \sqrt{2}\), where buckling occurs in one-half buckling wave. When \(a/b\) is greater than \(\sqrt{2}\) the buckling mode changes into multi-half buckling waves depending on \(a/b\) and \(\sigma_x/\sigma_y\).

For any aspect ratio, \(a/b\), the number of half buckling waves decreases as \(\sigma_x/\sigma_y\) decreases until it reaches one-half buckling wave for small values of \(\sigma_x/\sigma_y\).

The interaction relationship in this case is represented by a group of intersecting straight lines, as shown in Fig. 2. Each straight line corresponds to a buckling mode and the intersection points (knuckles) indicate transition of the buckling mode.

As \(a/b\) increases, the buckling interaction relationship between \(\sigma_x\) and \(\sigma_y\) becomes more convex with more knuckles on it, as shown in Fig. 2. This also implies that the interaction between \(\sigma_x\) and \(\sigma_y\) becomes less pronounced. If \(\sigma_x\) and \(\sigma_y\) are applied to a rectangular plate, the exact solution may be used to represent the buckling interaction relationship. However, a continuous equation is preferable in order to derive an interaction equation for general load components.
Although it is difficult to express accurately the exact solution for all $a/b$ ratios by a single continuous equation, the following equation gives a good, rather slightly conservative, approximation of the exact solution, as shown by the dotted lines in Fig. 2.

\[
\left( \frac{\sigma_x}{\sigma_{xcx}} \right)^{a_1} + \left( \frac{\sigma_y}{\sigma_{ycr}} \right)^{a_2} - 1 = 0
\]  

(14)

where,

\[
\begin{align*}
\alpha_1 &= a_2 = 1.0 \\
\alpha_2 &= 1.0 \\
\beta_1 &= 1.110\beta - 0.569 \\
\beta_2 &= 0.45\beta + 0.10 \\
\beta_3 &= 0.125\beta + 1.40 \\
\beta_4 &= -0.20\beta^2 - 1.60\beta + 5.10 \\
\beta_5 &= -0.30\beta^2 + 3.10 \\
\beta_6 &= 5.10 \\
\beta_7 &= 0.70 \\
\beta &= a/b
\end{align*}
\]

(2) Longitudinal compression $\sigma_x$ and longitudinal bending $\sigma_{bx}$

Figure 3 shows the buckling interaction relationships of plates with different aspect ratios subjected to axial compression $\sigma_x$ in the $x$ direction and in-plane bending $\sigma_{bx}$ in the same direction. These relations are obtained using the energy method. Although the solution in this case is approximate (upper bound), it should be quite accurate because only a few terms (2 or 3 terms) of the deflection series are required for convergence.

As may be seen from Fig. 3, this relationship is insensitive to the $a/b$ ratios and may be represented accurately by the following equation, which is also plotted by a dotted line in the figure.

\[
\frac{\sigma_x}{\sigma_{xcx}} + \left( \frac{\sigma_{bx}}{\sigma_{bxcr}} \right)^2 - 1 = 0
\]  

(15)

(3) Transverse compression $\sigma_y$ and longitudinal bending $\sigma_{bx}$

The buckling interaction relationships in this case are also obtained by the energy method and presented in Fig. 4. As in case (2), only a few terms of the deflection series are necessary to obtain convergence and the solutions may be quite accurate. These relations are well approximated by the following equation (see Fig. 4):

\[
\left( \frac{\sigma_y}{\sigma_{ycr}} \right)^{a_3} + \left( \frac{\sigma_{bx}}{\sigma_{bxcr}} \right)^{a_4} - 1 = 0
\]  

(16)

where,

\[
\begin{align*}
\alpha_3 &= \alpha_4 = 1.50\beta - 0.30 \\
1 \leq \beta \leq 1.6
\end{align*}
\]
\( \alpha_3 = 0.625\beta + 3.10 \)  
\( \alpha_4 = 6.25\beta - 7.90 \)  
\( \alpha_5 = 1.1 \)  
\( \alpha_6 = 12.1 \)  
\( \{ 1.6 < \beta \leq 3.2 \} \)  
\( \{ 3.2 < \beta \} \)

(4) Longitudinal compression \( \sigma_x \) and transverse bending \( \sigma_{by} \)

The buckling load for small \( a/b \) ratios may be calculated accurately using 3 terms of the deflection series. As \( a/b \) increases, more terms would be necessary to obtain accurate solutions by the energy method.

The buckling interaction relationships presented in Fig. 5 for \( a/b = 1 \) and 1.25 are calculated by the energy method and those for \( a/b = 3 \) and 5 by the incremental Galerkin method using a larger number of terms of the deflection series (7 to 11 terms).

Fig. 5 Buckling interaction between \( \sigma_x \) and \( \sigma_{by} \)

Convergence has been checked and ensured. These solutions are believed to be highly accurate. For \( a/b = \infty \) the problem handles biaxial compressions since the effect of the short boundaries vanishes. In this case, the interaction relationship for biaxial compressions (Fig. 2) is adopted.

The following equation yields a good approximation to these relationships as may be seen in Fig. 5.

\[ \left( \frac{\sigma_x}{\sigma_{xcr}} \right)^\eta_x + \left( \frac{\sigma_{by}}{\sigma_{bycr}} \right)^\eta_y - 1 = 0 \]  

(17)

where,

\( \alpha_7 = 0.930\beta^2 - 2.890\beta + 3.160 \)  
\( \alpha_6 = 1.20 \)  
\( \alpha_5 = 0.066\beta^2 - 0.246\beta + 1.328 \)  
\( \alpha_6 = 1.20 \)  
\( \alpha_5 = 1.117\beta - 3.837 \)  
\( \alpha_6 = -0.167\beta + 2.035 \)  
\( \{ 1 \leq \beta \leq 2 \} \)

\( \{ 2 < \beta \leq 5 \} \)

\( \{ 5 < \beta \leq 8 \} \)

\( \alpha_8 = 5.10 \)  
\( \alpha_6 = 0.70 \)  
\( \{ 8 < \beta \} \)

(5) Transverse compression \( \sigma_y \) and transverse bending \( \sigma_{by} \)

The buckling shape in this case may be represented accurately by a few terms of the deflection series. The energy method is used with enough accuracy in deriving the buckling interaction relationships for \( a/b = 1, 1.25 \) and 5. These relationships are shown in Fig. 6.

When \( a/b \) approaches infinitely, \( \sigma_{by} \) may be replaced by transversely uniform compression as stated in (4). The relationship between \( \sigma_y \) and \( \sigma_{by} \) becomes a straight line as shown in Fig. 6.

The following equation yields a good approximation to these relationships as may be seen in Fig. 6.

\[ \left( \frac{\sigma_y}{\sigma_{ycr}} \right)^\eta_x + \left( \frac{\sigma_{by}}{\sigma_{bycr}} \right)^\eta_y - 1 = 0 \]  

(18)

where,

\( \alpha_7 = 1.0 \)  
\( \alpha_8 = 14.0 - \beta \)  
\( \alpha_7 = 1.0 \)  
\( \alpha_8 = 1.0 \)  
\( \{ 1 \leq \beta \leq 7.5 \} \)

\( \{ 7.5 < \beta \} \)

Fig. 6 Buckling interaction between \( \sigma_y \) and \( \sigma_{by} \)

(6) Biaxial bending \( \sigma_{bx} \) and \( \sigma_{by} \)

The buckling shape associated with this load combination requires many terms of the deflection series in order to obtain accurate solutions. It is not appropriate to use the energy method in this case.

The interaction relationships presented in Fig. 7 are obtained by the incremental Galerkin method using 9 to 22 terms of the deflection series with convergence checked and ensured.
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![Graph showing buckling interaction between $\sigma_{bx}$ and $\sigma_{by}$](image)

Fig. 7 Buckling interaction between $\sigma_{bx}$ and $\sigma_{by}$

These relations may be well approximated by the following equation as may be seen in Fig. 7:

$$
\left( \frac{\sigma_{bx}}{\sigma_{bxCr}} \right)^{\alpha_{9}} + \left( \frac{\sigma_{by}}{\sigma_{byCr}} \right)^{\alpha_{10}} = 1 = 0
$$  (19)

where,

$\alpha_9 = 0.050\beta + 1.080$

$\alpha_{10} = 0.268\beta - 1.248/\beta + 2.112$

$\alpha_9 = 0.146\beta^2 - 0.533\beta + 1.515$

$\alpha_{10} = 0.268\beta - 1.248/\beta + 2.112$

$\alpha_9 = 5.0\beta - 13.50$

$\alpha_{10} = -0.70\beta + 6.70$

$\alpha_9 = 12.10$

$\alpha_{10} = 1.10$

(7) Longitudinal compression $\sigma_x$ and uniform shear $\tau$

Figures 8 to 11 represent the buckling interaction relationships between shear $\tau$ and each of the other four load components considered in this paper. Solutions, under the influence of shear stress, require much more terms of the solution series to obtain convergence specially for higher aspect ratios.

Solutions presented in these figures are obtained by the energy method. They usually tend to be less accurate than the cases of normal stresses. However, they are considered to have reasonable accuracy for practical purposes. Although available solutions do not cover the whole range of $a/b$ ratios, they cover the more used range.

These relationships may be approximated reasonably and rather conservatively by the following equations, which are plotted in respective figures.

The buckling interaction relationship between longitudinal compression $\sigma_x$ and uniform shear $\tau$ is expressed as follows (see Fig. 8):

$$
\frac{\sigma_x}{\sigma_{xcr}} + \left( \frac{\tau}{\tau_{cr}} \right)^{\alpha_{11}} = 1 = 0
$$  (20)

where,

$\alpha_{11} = -0.160\beta^2 + 1.080\beta + 1.082$ \quad $1 \leq \beta \leq 3.2$

$\alpha_{11} = 2.90$ \quad $3.2 < \beta$

(8) Transverse compression $\sigma_y$ and uniform shear $\tau$

In the same manner as stated in (7), the buckling equation between transverse compression $\sigma_y$ and uniform shear $\tau$ is expressed as (see Fig. 9):

$$
\frac{\sigma_y}{\sigma_{ycr}} + \left( \frac{\tau}{\tau_{cr}} \right)^{\alpha_{12}} = 1 = 0
$$  (21)

where,

$\alpha_{12} = 0.10\beta + 1.90$ \quad $1 \leq \beta \leq 2$

$\alpha_{12} = 0.70\beta + 0.70$ \quad $2 < \beta \leq 6$

$\alpha_{12} = 4.90$ \quad $6 < \beta$

(9) Longitudinal bending $\sigma_{bx}$ and uniform shear $\tau$

As may be seen from Fig. 10, the interaction relation is

![Graph showing buckling interaction between $\sigma_{by}$ and $\tau$](image)

![Graph showing buckling interaction between $\sigma_{by}$ and $\tau$](image)

Fig. 8 Buckling interaction between $\sigma_x$ and $\tau$

Fig. 9 Buckling interaction between $\sigma_y$ and $\tau$

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insensitive to the $a/b$ ratio and may be approximated accurately by the following equation, which is also plotted in the figure.

\[
\left( \frac{\sigma_{by}}{\sigma_{bxcr}} \right)^2 + \left( \frac{\tau}{\tau_{cr}} \right)^2 - 1 = 0
\]  

(22)

(10) **Transverse bending $\sigma_{by}$ and uniform shear $\tau$**

In case of transverse in-plane bending $\sigma_{by}$ and uniform shear $\tau$, which may be an important combination in design of deep girder, a series of eigen value analysis for the plate with $a/b = 3$ is carried out by the finite element method to extend the available range of $a/b$ ratio.

However, these relationships may be insensitive to $a/b$ ratio, and may be approximated reasonably by the following equation (see Fig. 11).

\[
\left( \frac{\sigma_{by}}{\sigma_{bxcr}} \right)^2 + \left( \frac{\tau}{\tau_{cr}} \right)^2 - 1 = 0
\]  

(23)

3.2 Buckling interaction for five load components

A general interaction relationship among the five load components is derived on the basis of the interaction relationships between two load components as presented in sec. 3.1.

At the first step, a set of two interaction relationships between two load components are chosen so that one of the load components is common in both relationships. The two relationships are combined to obtain, a new relationship among three load components.

By carrying on this procedure furthermore, the interaction relationship among the five load components is obtained.

In the following, the procedure for finding out the interaction among five load components is presented in detail.

(1) **Interaction among three load components**

Equations (14), (15) and (16) are respectively the interactions between $\sigma_x$ and $\sigma_y$, $\sigma_x$ and $\sigma_{bx}$, and $\sigma_y$ and $\sigma_{bx}$. Making combination of these relationships, an interaction relationship among three load components, $\sigma_x$, $\sigma_y$ and $\sigma_{bx}$ is first derived. Its schematic representation is shown in Fig. 12.

Let's consider the plate that buckles under these three load components which are denoted by $\sigma_x$, $\sigma_y$ and $\sigma_{bx}$, respectively.

When no transverse compression is applied $\sigma_y = 0$, the interaction among $\sigma_x$, $\sigma_{bx}$ and $\sigma_y$, corresponds to that between $\sigma_x$ and $\sigma_{bx}$, Eq.(15). In this case, the critical value $\sigma_y^{cr}$ of the longitudinal compressive stress $\sigma_y$ which together with $\sigma_{bx}$ causes buckling of the plate may be obtained by Eq.(15) as follows:
\[ \sigma_x^* = \sigma_{xcr} \left[ 1 - \left( \frac{\sigma_{by}}{\sigma_{bycr}} \right)^2 \right] \]  \hspace{1cm} (24)

\[ \sigma_y^* \] is represented by point A in the \( S_x - M_x \) plane in Fig. 12. Similarly, when no longitudinal compression is applied, \( \sigma_x = 0 \), the transverse compressive stress \( \sigma_y^* \) which causes buckling of the plate may be obtained by Eq.(16).

\[ \sigma_y^* = \sigma_{ycr} \left[ 1 - \left( \frac{\sigma_{by}}{\sigma_{bycr}} \right)^2 \right] \] \hspace{1cm} (25)

\( \sigma_y^* \) is represented by point B in the \( S_y - M_y \) plane in Fig. 12. It may be assumed that a similar relation to that of Eq.(14) exists between \( \sigma_x \) and \( \sigma_y \) in any plane of \( M_x = \sigma_{bx}/\sigma_{bxcr} = \) constant.

Replacing \( \sigma_{xcr} \) and \( \sigma_{ycr} \) in Eq.(14) by \( \sigma_x^* \) and \( \sigma_y^* \) respectively, the following buckling interaction relationship among \( \sigma_x, \sigma_y, \) and \( \sigma_{bx} \) may be obtained;

\[ \left( \frac{\sigma_x}{C_1 \sigma_{xcr}} \right)^{\alpha_1} + \left( \frac{\sigma_y}{C_2 \sigma_{ycr}} \right)^{\alpha_2} - 1 = 0 \] \hspace{1cm} (26)

where,

\[ C_1 = 1 - \left( \frac{\sigma_{bx}}{\sigma_{bxcr}} \right)^2 \]
\[ C_2 = \left[ 1 - \left( \frac{\sigma_{bx}}{\sigma_{bxcr}} \right)^2 \right]^{-\frac{1}{\alpha_2}} \]

(2) Interaction among four load components

In a similar way, Eqs.(17), (18) and (19) may be combined with Eq.(26) to obtain a relationship among \( \sigma_x, \sigma_y, \sigma_{bx} \) and \( \sigma_{by} \) as follows:

Under any value of \( \sigma_x^* \), which is smaller than \( \sigma_{bycr} \), the longitudinal compression \( \sigma_y^* \) which causes buckling of the plate may be obtained by Eq.(17) when \( \sigma_y = \sigma_{bx} = 0 \).

\[ \sigma_x^* = \sigma_{xcr} \left[ 1 - \left( \frac{\sigma_{bx}}{\sigma_{bycr}} \right)^2 \right] \] \hspace{1cm} (27)

Under the same value of \( \sigma_x^* \), the transverse compression \( \sigma_y^* \) which causes buckling of the plate may also be obtained by Eq.(18) when \( \sigma_x = \sigma_y = 0 \)

\[ \sigma_y^* = \sigma_{ycr} \left[ 1 - \left( \frac{\sigma_{by}}{\sigma_{bycr}} \right)^2 \right] \] \hspace{1cm} (28)

Under the same value of \( \sigma_y^* \), the longitudinal bending \( \sigma_{bx}^* \) which causes buckling of the plate may be obtained by Eq.(19) when \( \sigma_x = \sigma_y = 0 \)

\[ \sigma_{bx}^* = \sigma_{bxcr} \left[ 1 - \left( \frac{\sigma_{by}}{\sigma_{bycr}} \right)^2 \right] \] \hspace{1cm} (29)

\( \sigma_x^*, \sigma_y^* \) and \( \sigma_{bx}^* \) in the above equations are represented in Fig. 12 and they are substituted in \( \sigma_{xcr}, \sigma_{ycr} \) and \( \sigma_{xcr} \) in Eq.(26), respectively.

Assuming that the relation among \( \sigma_x, \sigma_y, \sigma_{bx} \) for \( \sigma_y^* \) is constant is similar to that for \( \sigma_{by} = 0 \). The following interaction relationship among \( \sigma_x, \sigma_y, \sigma_{bx} \) and \( \sigma_{by} \) may be obtained by substituting \( \sigma_x^*, \sigma_y^* \) and \( \sigma_{bx}^* \) for \( \sigma_{xcr}, \sigma_{ycr} \) and \( \sigma_{bxcr} \) in Eq.(26).

\[ \left( \frac{\sigma_x}{C_3 \sigma_{xcr}} \right)^{\alpha_1} + \left( \frac{\sigma_y}{C_5 \sigma_{ycr}} \right)^{\alpha_2} - 1 = 0 \] \hspace{1cm} (30)

where,

\[ C_3 = \left[ 1 - \left( \frac{\sigma_{by}}{\sigma_{bycr}} \right)^2 \right]^{-\frac{1}{\alpha_1}} \]
\[ C_4 = \left[ 1 - \left( \frac{\sigma_{bx}}{\sigma_{bxcr}} \right)^2 \right]^{-\frac{1}{\alpha_1}} \]
\[ C_5 = \left[ 1 - \left( \frac{\sigma_{bx}}{\sigma_{bycr}} \right)^2 \right]^{-\frac{1}{\alpha_2}} \]
\[ C_6 = \left[ 1 - \left( \frac{\sigma_{bx}}{\sigma_{xcrcr}} \right)^2 \right]^{-\frac{1}{\alpha_2}} \]
\[ C_7 = \left[ 1 - \left( \frac{\sigma_{bx}}{\sigma_{bycr}} \right)^2 \right]^{-\frac{1}{\alpha_2}} \]

(3) Interaction among five load components

Finally, Eqs.(20), (21), (22) and (23) are combined with Eq.(30) as follows:

Under any constant value of uniform shear \( \tau_{y**} \) which is smaller than \( \tau_{cr} \), \( \sigma_y^* \) which causes buckling of the plate may be obtained by Eq.(20) when the other load components vanish;

\[ \sigma_y^* = \sigma_{ycr} \left[ 1 - \left( \frac{\tau_{y**}}{\tau_{cr}} \right)^{\alpha_2} \right] \] \hspace{1cm} (31)

Similarly, \( \sigma_x^*, \sigma_{bx}^* \) may be obtained by Eqs.(21), (22) and (23) respectively.

\[ \sigma_x^* = \sigma_{xcr} \left[ 1 - \left( \frac{\tau_{x**}}{\tau_{cr}} \right)^{\alpha_1} \right] \] \hspace{1cm} (32)

\[ \sigma_{bx}^* = \sigma_{bxcr} \left[ 1 - \left( \frac{\tau_{x**}}{\tau_{cr}} \right)^{\alpha_1} \right]^{0.5} \] \hspace{1cm} (33)

\[ \sigma_{by}^* = \sigma_{bycr} \left[ 1 - \left( \frac{\tau_{y**}}{\tau_{cr}} \right)^{\alpha_2} \right]^{0.5} \] \hspace{1cm} (34)

Assuming that the interaction among \( \sigma_x, \sigma_y, \sigma_{bx}, \) and \( \sigma_{by} \) for any value of shear \( \tau_{y**} = \) constant is similar to that for \( \tau = 0 \), the following interaction equation is obtained by substituting \( \sigma_x^*, \sigma_y^*, \sigma_{bx}^* \) and \( \sigma_{by}^* \) for \( \sigma_{xcr}, \sigma_{ycr}, \sigma_{bxcr} \) and \( \sigma_{bycr} \) respectively into Eq.(30).

\[ \Gamma_B = (C_8 C_9 C_{10})^{\alpha_1} (C_{11} C_{12} C_{13})^{\alpha_2} \left[ \left( \frac{\sigma_x}{C_6 C_8 C_9} \sigma_{xcr} \right)^{\alpha_1} + \left( \frac{\sigma_y}{C_7 C_{10} C_{12}} \sigma_{ycr} \right)^{\alpha_2} - 1 \right] \] \hspace{1cm} (35.a)

where,

\[ C_8 = 1 - \left( \frac{\tau_{y**}}{\tau_{cr}} \right)^{\alpha_1} \]
\[ C_9 = \left[ 1 - \left( \frac{\sigma_{by}}{C_{14} \sigma_{bycr}} \right)^{2} \right]^{1/\alpha_1} \]
\[ C_{10} = 1 - \left( \frac{\sigma_{bx}}{C_{14} \sigma_{bxcr}} \right)^{2} \]
\[ C_{11} = 1 - \left( \frac{\tau_{x**}}{\tau_{cr}} \right)^{\alpha_1} \]
\[ C_{12} = \left[ 1 - \left( \frac{\sigma_{by}}{C_{14} \sigma_{bycr}} \right)^{2} \right]^{1/\alpha_1} \]
\[ C_{13} = \left[ 1 - \left( \frac{\sigma_{bx}}{C_{14} \sigma_{bxcr}} \right)^{2} \right]^{1/\alpha_2} \]
\[ C_{14} = 1 - \left( \frac{\tau_{y**}}{\tau_{cr}} \right)^{\alpha_2} \]
\[ C_{15} = \left[ 1 - \left( \frac{\sigma_{by}}{C_{14} \sigma_{bycr}} \right)^{2} \right]^{1/\alpha_2} \]

Consequently, the buckling interaction equation among five load components is obtained as follows:

\[ \Gamma_B = 0 \] \hspace{1cm} (35.b)

Coefficients \( \alpha_1 \) to \( \alpha_{12} \) included in the above equation are given with Eqs.(14) to (21). If \( \Gamma_B \) is smaller than zero with applied stresses substituted into Eq.(35), the plate does not buckle. If the applied stresses are gradually increased until they satisfy Eq.(35.b), the plate buckles.
3.3 Method of solution of proposed equation

At first sight, Eq.(35) seems somewhat complicated to solve analytically. However, it is not difficult to solve it numerically. Procedure of solution may differ according to the particular problem in hand, such as finding out the buckling stresses in case of proportionally increasing load, or finding out the critical value of a load component which increases gradually while other load components are kept constant, or finding out the critical thickness of the plate under a given load, etc.

However, in the following, general procedure is proposed to solve a general problem. Many other problems may be treated in a similar way.

(a) Definition of problem

As shown in Fig. 1, a simply supported rectangular plate \((a \times b \times t)\) is subjected to a set of initial stresses composed of \(\sigma_x^o, \sigma_y^o, \sigma_{bx}^o, \sigma_{by}^o\), \(\tau^o\) and a gradually increasing load which produces additional stress \(\eta\sigma_x, \eta\sigma_y, \eta\sigma_{bx}, \eta\sigma_{by}, \eta\tau\), where \(\eta\) is a load factor.

It is required to find out the critical value of load factor \(\eta\) with which the plate just buckles. In this problem, by setting non-existing load components at zero or equating the initial load components to zero, a wide variety of problems may be formulated.

(b) Method of solution

For any value of \(\eta\), the stresses acting on the plate \((\sigma_x, \sigma_y, \sigma_{bx}, \sigma_{by}, \tau)\) may be calculated as follows:

\[
\begin{align*}
\sigma_x &= \sigma_x^o + \eta\sigma_x \\
\sigma_y &= \sigma_y^o + \eta\sigma_y \\
\sigma_{bx} &= \sigma_{bx}^o + \eta\sigma_{bx} \\
\sigma_{by} &= \sigma_{by}^o + \eta\sigma_{by} \\
\tau &= \tau^o + \eta\tau
\end{align*}
\]

(36)

The solution may proceed as follows:

1. Evaluate first the pure buckling stress such as \(\sigma_{xcr}, \sigma_{ycr}, \sigma_{bxcr}, \sigma_{bycr}\) and \(\tau_{cr}\), when a rectangular plate is subjected to a single load (See Appendix)

2. Compute the values of \(\alpha_1\) to \(\alpha_{12}\)

3. Substitute the initial load into Eq.(35) and calculate \(\Gamma_B\) to see that the plate does not buckle under the initial load (\(\Gamma_B\) is negative)

4. Guided by \(\Gamma_B\) obtained in 3, assume two or three values of load factor, \(\eta_1, \eta_2\) and \(\eta_3\), substitute them in Eq. (36) and put the resulting stresses corresponding to each \(\eta\) into Eq.(35). Then, calculate values of \(\Gamma_B\).

5. Plot \(\Gamma_B\) against \(\eta\) as shown in Fig. 13 and find out the value of \(\eta\) at which \(\Gamma_B\) becomes equal to zero. This \(\eta\) may be denoted by \(\eta_{cr}\) and is the critical value of the load factor when the plate just buckles.

4. Assessment of Accuracy of Proposed Interaction Equation

4.1 Buckling interaction equation for two load components

As shown above, a buckling interaction equation is derived for a simply supported rectangular plate subjected to five load components. From this equation, Eq.(35), the interaction equation between each two load components, Eqs.(14) to (23) can easily be derived.

As may be seen in Figs. 2 to 11, these equations are accurate enough to be applied to practical design. For infinitely long plates, \(a/b = \infty\), the buckling interactions seem to change suddenly, therefore the accuracy of the new proposed interaction equation may be reduced. However, for practical aspect ratios, the accuracy is good enough and the estimated buckling load tends to be conservative.

4.2 Buckling interaction equation for three load components

Further assessment of accuracy is carried out for cases with three load components, with four or five load components.

As mentioned above, it was assumed that the interaction relationship among three load components is similar to that between two load components, as shown in Fig. 12. This assumption does not have much effect on accuracy for practical aspect ratios of the plate and a similar accuracy may be expected as shown later.

In the following, the accuracy of buckling load subjected to combined three load components is investigated by comparison between estimated values, Eq.(35) and values calculated by either the incremental Galerkin method or the energy method.

(a) Biaxial compressions \(\sigma_x\) and \(\sigma_y\) and shear \(\tau\)

Figure 14 compares the buckling load of a rectangular
plate with different aspect ratios estimated by Eq.(35) and that calculated by the incremental Galerkin method for a plate subjected to biaxial compression and shear, of which the load ratio is $\sigma_x / \sigma_{xcr} = \sigma_y / \sigma_{ycr} = \tau / \tau_{cr}$. They show good agreement.

Fig. 14 Comparison between proposed equation, Eq.(35) and theoretical buckling load of rectangular plates subjected to $\sigma_x$, $\sigma_y$ and $\tau$.

(b) Biaxial compressions $\sigma_x$ and $\sigma_y$ and longitudinal bending $\sigma_{bx}$

Figures 15.a, b and c show the buckling interaction relationships of rectangular plates with different aspect ratios subjected to biaxial compressions and in-plane bending in the longitudinal direction, of which the load ratios $\sigma_{bx} / \sigma_{bxcr}$ with respect to $\sigma_y / \sigma_{ycr}$ are varied from 0.5, 1.0 to 2.0. The results by Eq.(35) are compared with those by the energy method. The buckling loads estimated by Eq.(35) are seen to be accurate enough.

(c) Transverse compression $\sigma_y$, longitudinal bending $\sigma_{bx}$ and shear $\tau$

The buckling interaction relationship of an infinitely long plate subjected to transverse compression, longitudinal in-plane bending and shear is presented in Figs. 16.a and b.

In Fig. 16.a, $\tau / \tau_{cr} = \sigma_{bx} / \sigma_{bxcr}$ relationship are presented for two cases of constant transverse compression, of which $\sigma_y / \sigma_{ycr}$ is 0.5 and 0.8. On the other hand, the $\sigma_y / \sigma_{ycr}$ against $\tau / \tau_{cr}$ relationship are shown in Fig. 16.b for four cases of constant in-plane bending, of which $\sigma_{bx} / \sigma_{bxcr}$ is 0.0, 0.5, 0.8 and 0.9.

The solid lines show the result calculated by the energy method, while the dotted ones show those by Eq.(35). It may be seen from the above two figures that Eq.(35) is accurate enough for engineering purposes.

5. Elastic – Plastic Buckling

In the previous section, elastic buckling of the plate subjected to combined five load components has been discussed.

If the plate is perfectly flat, that is it has no initial imperfections, the plate will buckle elastically or plastically under uniform in-plane compression or uniform shear. Elastic – plastic buckling may not occur in this case but under in-plane bending, elastic – plastic buckling may take place in the plate because the plastic

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Fig. 15.a Comparison between proposed equation, Eq.(35) and theoretical buckling load of rectangular plates subjected to $\sigma_x$, $\sigma_y$ and $\sigma_{bx}$

Fig. 15.b Comparison between proposed equation, Eq.(35) and theoretical buckling load of rectangular plates subjected to $\sigma_x$, $\sigma_y$ and $\sigma_{bx}$

Fig. 15.c Comparison between proposed equation, Eq.(35) and theoretical buckling load of rectangular plates subjected to $\sigma_x$, $\sigma_y$ and $\sigma_{bx}$
Fig. 16.b Comparison between proposed equation, Eq. (35) and theoretical buckling interaction relationship of infinite plate subjected to $\sigma_y$, $\sigma_{bx}$ and $\tau$

Results of theoretical analyses of buckling and ultimate strengths of a rectangular plate are shown with respect to its slenderness ratio $\lambda = h/t\sqrt{\sigma_0/E}$, ($\sigma_0 =$ yield stress) in Fig. 179. The plate has different values of unital deflection and residual stresses.

According to Fig. 17, if the plate has no initial imperfections, it reaches ultimate limit state after elastic buckling or fails by plastic collapse.

In general, a plate with initial deflection and welding residual stress may not exhibit clear buckling phenomenon and reaches its ultimate strength state, as the load increases. Thin plates, in which the slenderness ratio is greater than about 2.4, the ultimate strength is higher than the buckling strength of a similar flat plate and the later is regarded as a failure condition.

On the other hand, in case of thicker plates, of which slenderness ratio is smaller than 2.4, initial imperfections cause a plate to fail by plastic collapse below elastic or plastic buckling strength. This type of behaviour is a phenomenon of plastic collapse. The nature of this behaviour is different from that arising in elastic buckling. Therefore, buckling strength is not likely to be valid for such thick plates as a failure condition.

From the above discussion, it may be better to treat plastic collapse phenomenon separately rather than to make corrections to the elastic buckling relationships. This paper concentrates on elastic buckling, while interaction relationships for plastic collapse are planned for future research.

6. Buckling Interaction Equations Proposed by Lloyd's Register and Det Norske Veritas and Their Accuracy

6.1 Comparison with an interaction equation proposed by Lloyd's Register

Lloyd's Register3 proposed the following buckling interaction equation for a simply supported rectangular plate subjected to five load components.

$$\left(\frac{\sigma_x}{\sigma_{scr}}\right)^2 + \left(\frac{\sigma_y}{\sigma_{ycr}}\right)^2 + \left(\frac{\sigma_{by}}{\sigma_{bycr}}\right)^2 + \left(\frac{\tau}{\tau_{cr}}\right)^2 = 1 \quad (37)$$

where,

$$\alpha = 0.6 + 0.4/\beta \quad 0.3 \leq \beta \leq 1.0$$

$$\alpha = 0.6 + 0.4/\beta \quad 1.0 \leq \beta \leq 3.5$$

$$\beta = a/b$$

It is should be noted that the above equation is valid for $1/3.5 \leq a/b \leq 3.5$. In case of two load components, this equation yields good approximation in some cases. However, in some other cases, it yields estimation on the non-conservative side of the theoretical relationships as
may be seen in Figs. 18.b, d, e and f.

Equation (35), proposed in this paper, is plotted in the same figures for comparison with Lloyd's equation. In case of in-plane bending and shear, both equations yield the same result as shown in Figs. 10 and 11. Eq.(35), however, yields better results for all other load combinations.

For combination of more than two load components, Lloyd's equation just adds nondimensional load components. Such direct addition may need justification.

6.2 Comparison with an interaction equation proposed by Det Norske Veritas

DnV\(^4\) proposed the following equation as a buckling interaction relationship for rectangular plates

\[
\eta = \sqrt{\eta^2_{cs} + 2\left(\frac{b}{a}\right)^2 \eta_{cs} \eta_c + \eta^2_c} \tag{38}
\]

where,

\(\eta\) = usage factor and when the factor of safety is considered to be more than 1.0, \(\eta\) is regarded as 1.0

\(\eta_{cs}\) = usage factor for the combination of shear and compression in one direction

For thin plates (1.4 < \(\lambda\), which is modified reduced slenderness ratio and defined in 3.3 of Ref.4),

\[
\eta_{cs} = \frac{1 + \psi}{4} \frac{\sigma}{\sigma_{cr}} + \left(\frac{\frac{3 - \psi}{4} \frac{\sigma}{\sigma_{cr}} + \left(\frac{\sigma}{\sigma_{cr}}\right)^2}{2}\right) \tag{39}
\]

where,

\(\eta_c\) = usage factor for compression in the other direction, i.e. \(\sigma/\sigma_{cr}\) in the other direction

\(\sigma\) = reference stress for compression in any one direction

\(\sigma_{cr}\) = critical value of the reference stress

\(\psi\) = stress multiplier, the ratio of stresses at plate edges as shown in Fig. 19 and is used to combine axial compression in one direction and in-plane bending in the same direction

DnV specifies values for \(\sigma_{cr}\) as follows:

\[
\sigma_{cr} = C \frac{\pi^2 E}{12(1 - \nu^2)} \left(\frac{t}{b}\right)^2 = C \sigma_E \tag{40}
\]

where, in the \(x\) direction,

\[C = \frac{8.4}{\psi + 1.1} \quad 0 < \psi \leq 1\]

\[C = 7.6 - 6.4\psi + 10\psi^2 - 1 \leq \psi < 0\]

and in the \(y\) direction,

\[C = \left[1 + \left(\frac{b}{a}\right)^2\right] \frac{2.1}{\psi + 1.1} \quad 0 < \psi \leq 1\]

\[C = (1 + \psi)C_a - \psi C_b + 10\psi(1 + \psi)\left(\frac{b}{a}\right)^2 \quad -1 < \psi < 0\]
Fig. 18.d Accuracy of buckling interaction equations for $\sigma_x$ and $\sigma_{by}$

Fig. 18.e Accuracy of buckling interaction equations for $\sigma_y$ and $\sigma_{by}$

$$C_a = 1.91 \left[1 + \left(\frac{b}{a}\right)^2\right]^2$$

$$C_b = 24\left(\frac{b}{a}\right)^2 \quad \frac{a}{b} \leq 1.5$$

$$C_b = 2 + 16\left(\frac{b}{a}\right)^2 + 8\left(\frac{b}{a}\right)^4 \quad \frac{a}{b} > 1.5$$

This relationships are also plotted in Figs. 18.a to h together with Eq.(35) proposed in this paper. Eq.(38) proposed by DnV makes accurate estimation of buckling strength in some cases. However, it underestimates buckling strength greatly in some other cases of complicated interaction.

7. Conclusion

In this paper, a new elastic buckling interaction equa-

Fig. 18.f Accuracy of buckling interaction equations for $\sigma_{bx}$ and $\sigma_{by}$

Fig. 18.g Accuracy of buckling interaction equations for $\sigma_x$ and $\tau$

Fig. 18.h Accuracy of buckling interaction equations for $\sigma_y$ and $\tau$
New Interaction Equation for Plate Buckling

\[
E = \text{modulus of elasticity}, \quad \nu = \text{Poisson's ratio}
\]

and coefficients \(k_x, k_y, k_{bx}, k_{by}, \) and \(k_{r}\) may be calculated for rectangular plates \(a/b \geq 1\) as follows:

\[
k_x = (m/\beta + \beta/m)^2
\]

\[
k_y = (1.0 + 1.0/\beta^2)^2
\]

\[
k_{bx} = 23.9
\]

\[
k_{by} = 23.9 \quad 1 \leq \beta \leq 1.5
\]

\[
= 15.87 + 1.87\beta^2 + 8.6/\beta^2 \quad \beta > 1.5
\]

\[
k_{r} = 5.34 + 4.0/\beta^2
\]

(approximate formula)

where,

\[
\beta = a/b
\]

\[
m = \text{buckling wave number in the } x \text{ direction and the minimum integer to satisfy the condition}
\]

\[
\beta \leq \sqrt{m(m+1)}
\]

References


3) LR PASS: Lloyd’s Register’s Plan Appraisal Systems for Ships.

4) DoV: Buckling Strength Analysis of Mobil Offshore Units, Det Norske Veritas, Classification Note No. 30.1, 1984.


